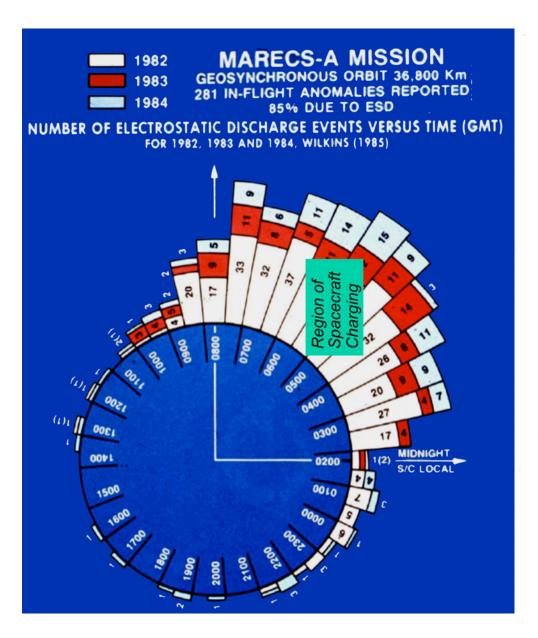
Spacecraft Charging Interactions

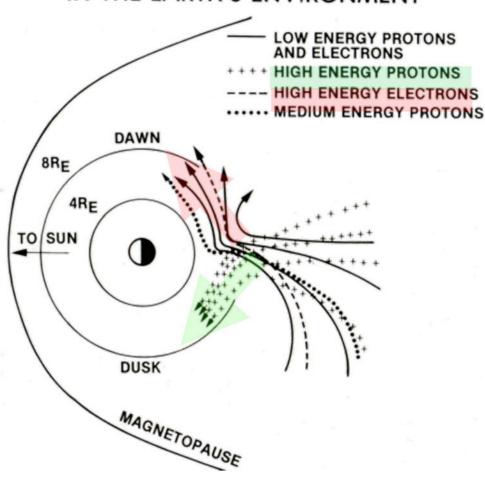
Dr. Henry B. Garrett
Jet Propulsion Laboratory
California Institute of Technolgy
Pasadena, CA 91109

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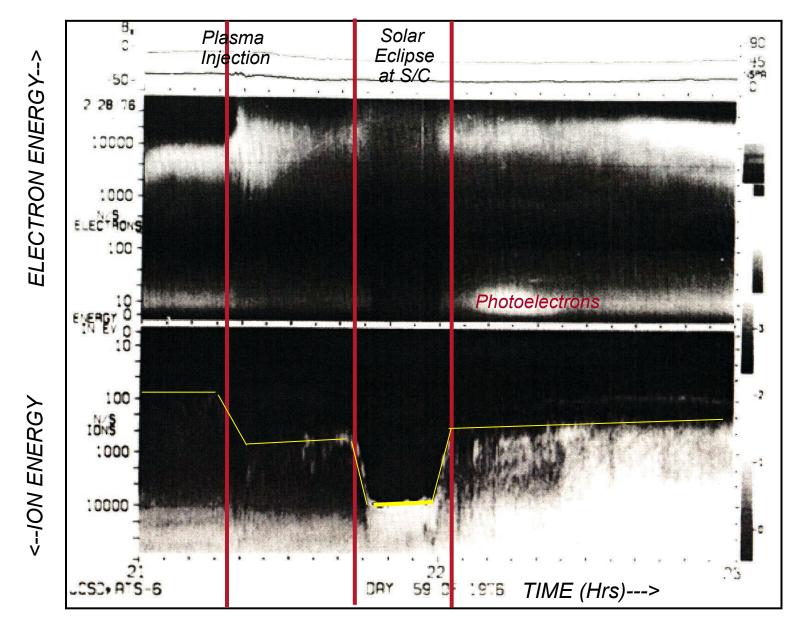
Plasma Environment—Surface Charging



PLASMA MOTIONS IN THE EARTH'S ENVIRONMENT



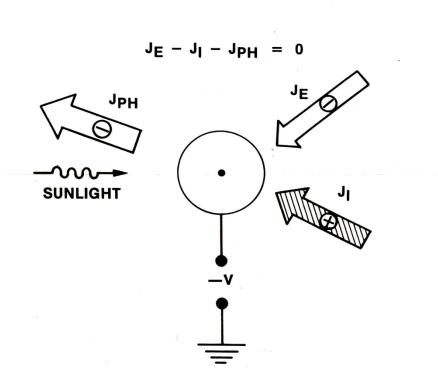
ATS-6 Spectrogram of Geosynchronous Charging

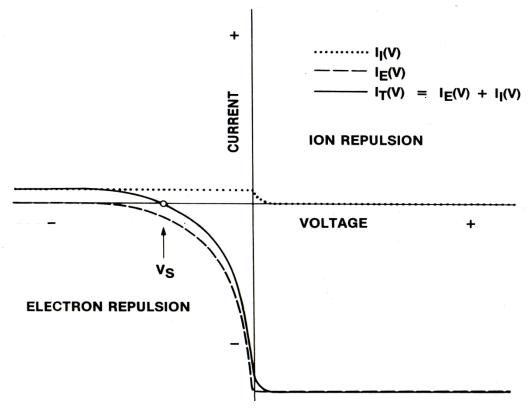


Theory of Spacecraft Charging: A Simple Picture ...

THEORY OF SPACECRAFT CHARGING

THEORY OF SPACECRAFT CHARGING





Theory of Spacecraft Charging: A Simple Example

FOR A NEGATIVELY CHARGED SPACECRAFT:

$$J_{T}(V) = J_{Io} \left(1 - \frac{qV}{KT_{I}} \right) - J_{eo} \left(e^{qV/KT_{e}} \right)$$

TYPICALLY AT GEOSYNCHRONOUS ORBIT:

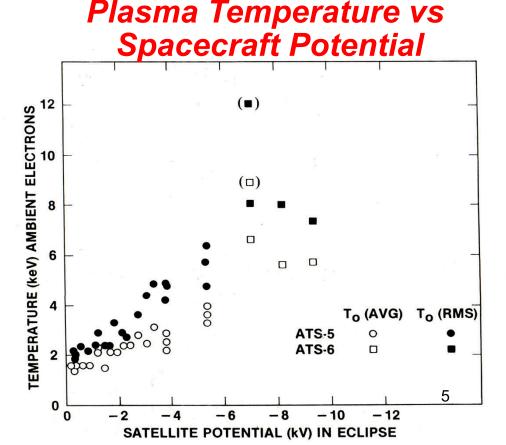
 $\frac{qV}{KT_I} \sim 0$

FOR CURRENT BALANCE:

$$J_T(V) = 0$$

THIS IMPLIES:

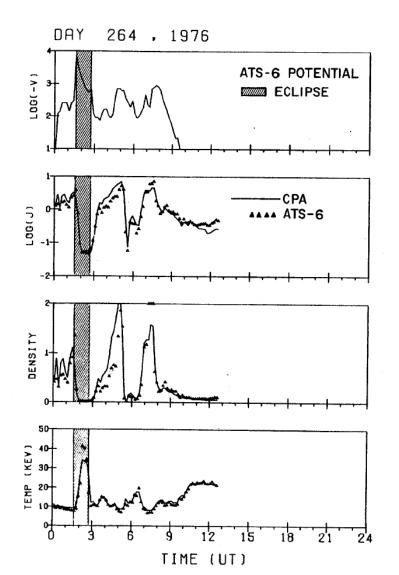
$$V = \frac{-KT_e}{q} \ln \left(\frac{J_{eo}}{J_{Io}} \right)$$

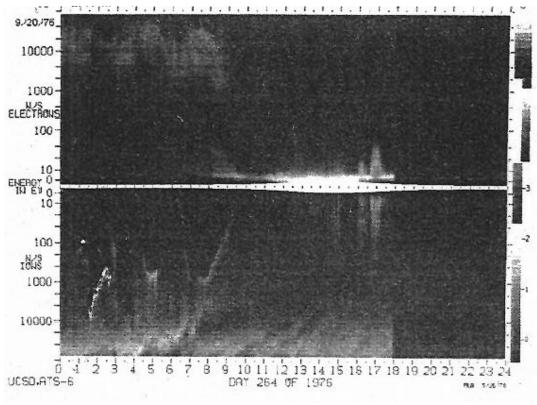


Electron Current versus Spacecraft Potential

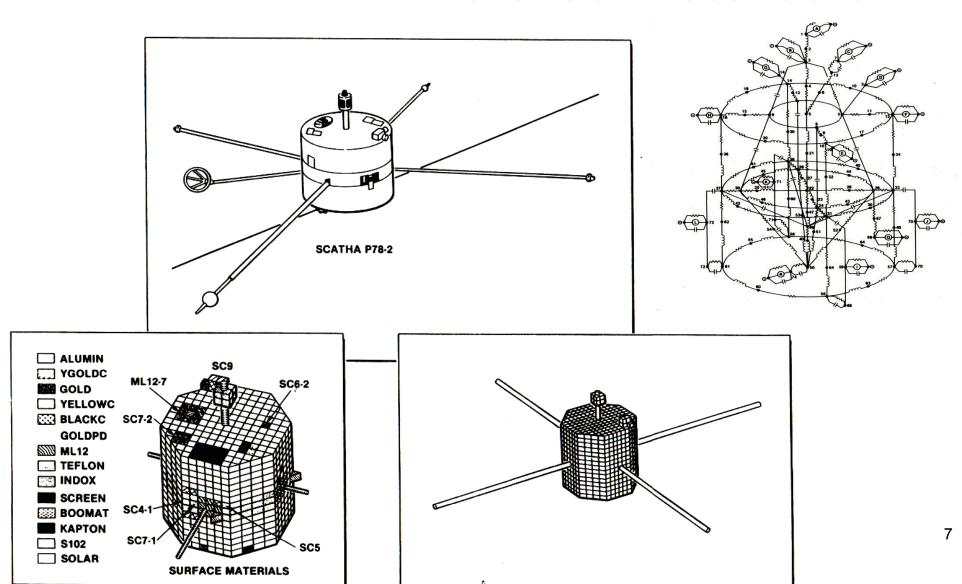
30-95 KeV ELECTRON DATA

ATS-6 SPECTROGRAM

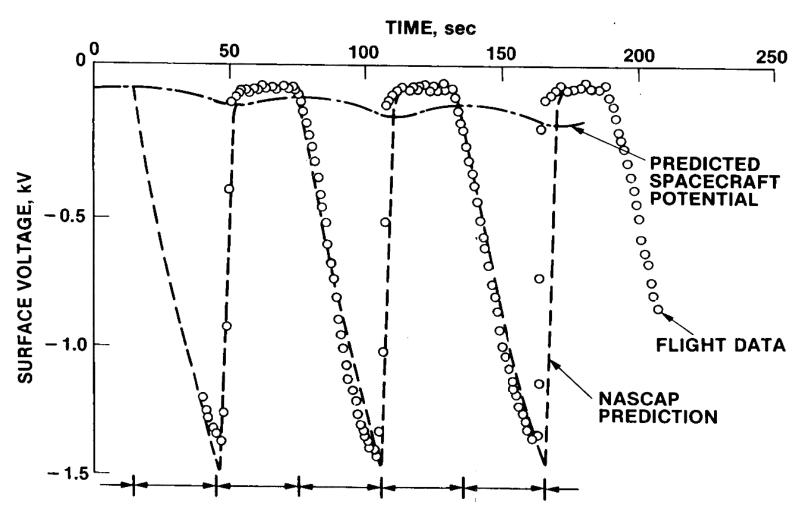




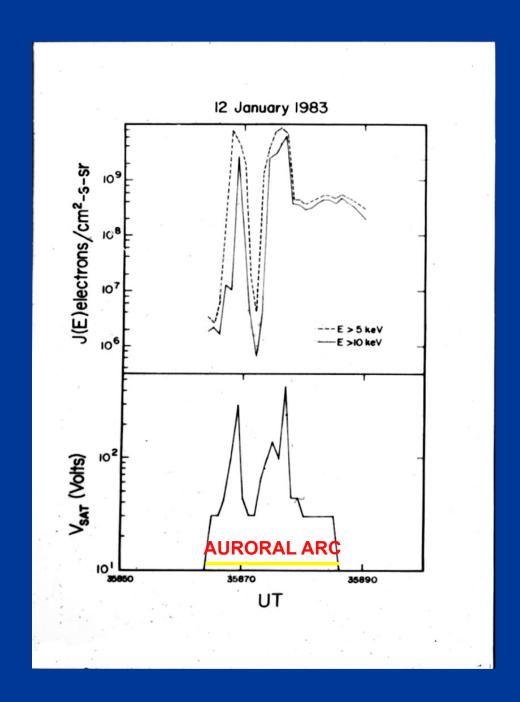
NASCAP Spacecraft Charging Code: Differential Potentials



Differential Potentials on SCATHA as Predicted by NASACP



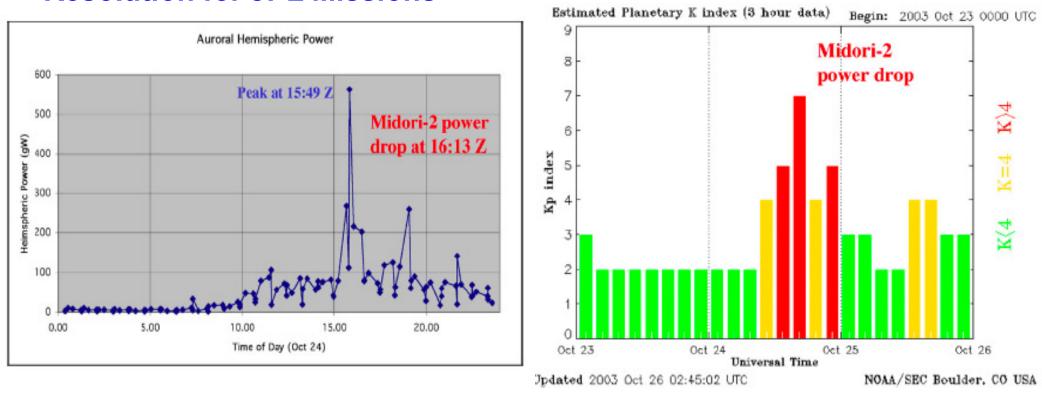
DMSP Low Altitude Spacecraft Charging





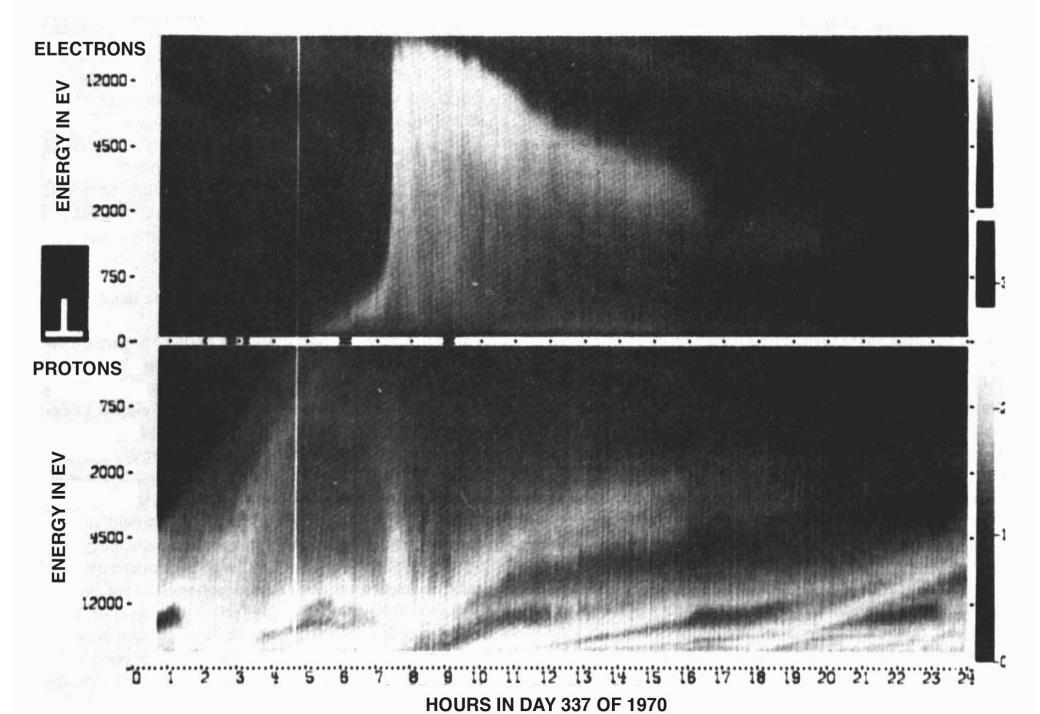
Auroral Effects on JPL Ops, Oct. 24, 2003

Lessons Learned: Geophysical Indices Critical to Rapid Anomaly Resolution for JPL Missions

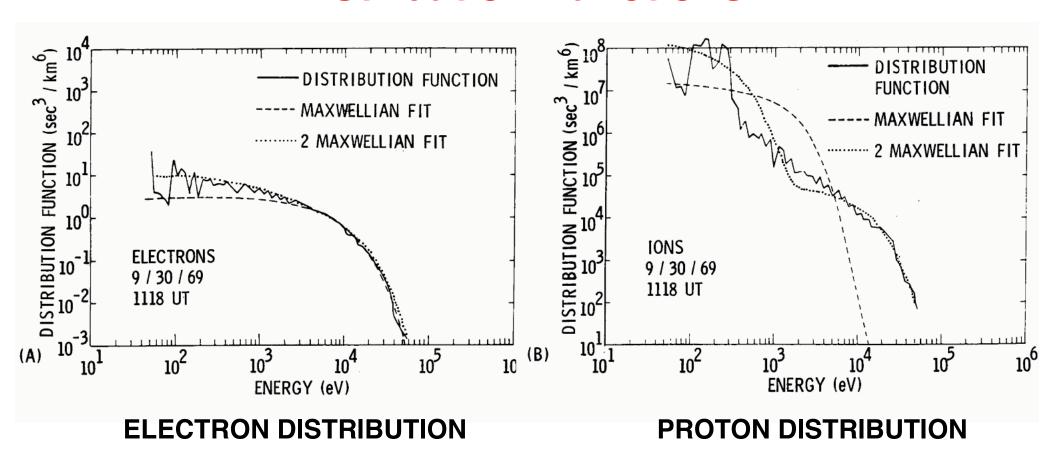


Oct 24: ADEOS-Midori-2 (JPL SeaWinds Instrument) Failed.
Attributed to Spacecraft Surface Charging

Space Weather: Spectrogram



Electron and Proton Geo Plasma Distribution Functions



Two Maxwellian Approach to Charging Environment

TWO MAXWELLIAN DISTRIBUTION FUNCTION

$$F_{2}(v) = \left(\frac{m}{2\pi}\right)^{3/2} \left[\frac{n_{1}}{\left(KT_{1}\right)^{3/2}} e^{\left(\frac{-mv^{2}}{2KT_{1}}\right)} + \frac{n_{2}}{\left(KT_{2}\right)^{3/2}} e^{\left(\frac{-mv^{2}}{2KT_{2}}\right)} \right]$$

TWO MAXWELLIAN PLASMA MOMENTS

NUMBER DENSITY: $M_1 = n_1 + n_2$

NUMBER FLUX: $M_2 = \frac{n_1}{2\pi} \left(\frac{2KT_1}{\pi m} \right)^{1/2} + \frac{n_2}{2\pi} \left(\frac{2KT_2}{\pi m} \right)^{1/2}$

ENERGY DENSITY: $M_3 = \frac{3}{2}n_1KT_1 + \frac{3}{2}n_2KT_2$

ENERGY FLUX: $M_4 = \frac{n_1 m}{2\pi} \left(\frac{2KT_1}{\pi m} \right)^{3/2} + \frac{n_2 m}{2\pi} \left(\frac{2KT_2}{\pi m} \right)^{3/2}$

GEOSYNCHRONOUS PLASMA MODEL

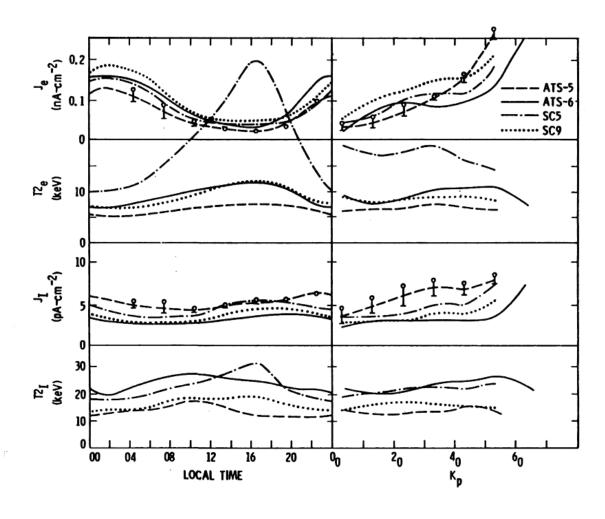
 $M_i(A_p, LT) = (a + bA_p)(c + d\cos[2\pi(LT - e)/24] + f\cos[4\pi(LT - g)/24])$

Geosynchronous 2 Maxwellian Statistics

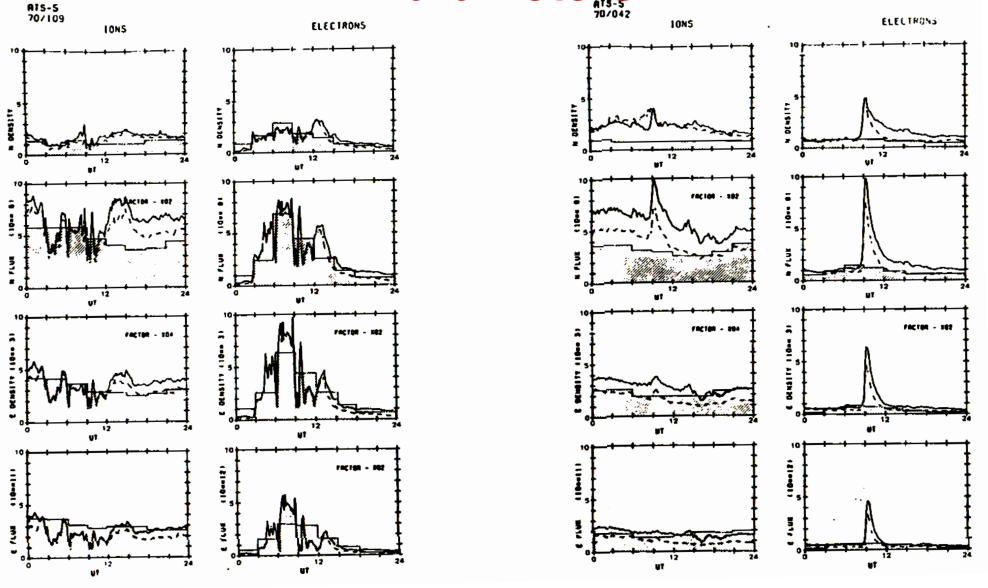
STATISTICAL DISTRIBUTIONS OF KEY GEOSYNCHRONOUS PARAMETERS

ELECTRON J (nA-cm⁻²) ION J (pA-cm²) ATS-5 % OCCURRENCE ATS-6 SC9 J_| (pA-cm⁻²) J_e (nA-cm⁻²) ELECTRON T2 (keV) ION T2 (keV) * OCCURRENCE T2_e (keV) 70 T2₁ (keV)

LOCAL TIME/Kp VARIATIONS FOR KEY GEOSYNCHRONOUS PLASMA PARAMETERS



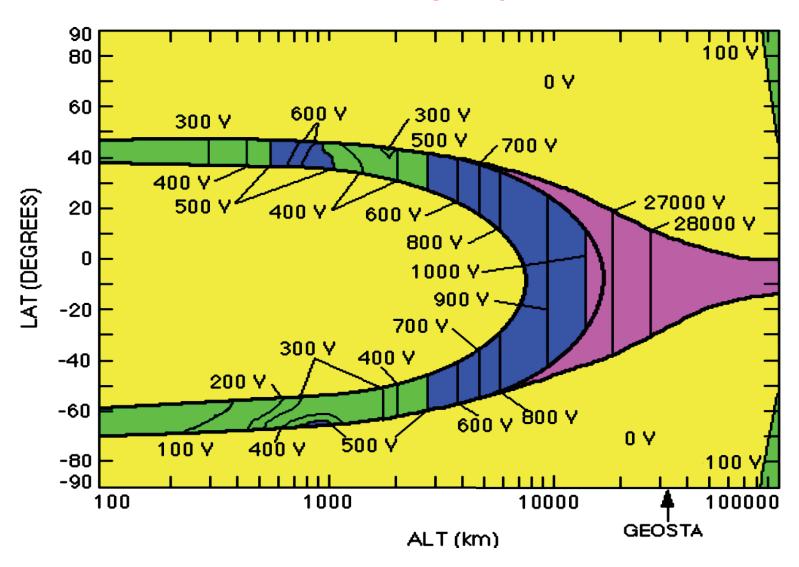
Modeled (Kp, LT) vs Observed Geo Plasma Parameters **THE TRUE TO BE THE PARAMETER SHOWN THE PARAMETER SHO



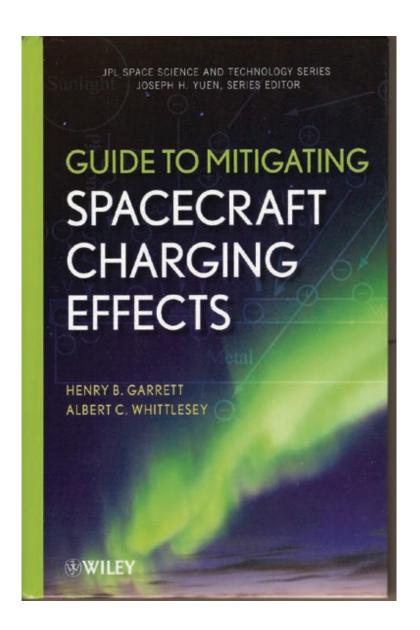
Summary:

- Surface Charging comes in 2 forms:
 - "Absolute" or "Spacecraft to Plasma Ground"
 - "Differential Charging" between surfaces
- Absolute Charging (10-30 KV) roughly proportional to the ambient electron current once a threshold temperature (~2-3 KV) is exceeded→Need Te and Je estimates
- Differential Charging is very spacecraft configuration dependent (spinner vs 3-axis, shadowed surfaces, surface properties, etc.)
- Absolute Charging can have little effect but indicative of possible high levels of Differential Charging
- Differential Charging source of arc discharges ($\Delta V < 100 V$?)
- Space Weather Forecasting useful for Absolute Charging but need spacecraft modeling (NASCAP-2K) for Differential Charging

Worst Case Surface Potentials for Earth Environment in the Absence of Sunlight (Evans et al., 1989)



Recommended Guide to Spacecraft Charging



Title: Guide to Mitigating Spacecraft Charging Effects

Authors: Henry B. Garrett and Albert C. Whittlesey

Publisher: Wiley, 2012

ISBN: 978-1-118-18645-9

BACKUP VIEWGRAPHS

"Analog" Model of Geo Orbit Environment

